



Designation: D 790 – 023

## Standard Test Methods for Flexural Properties of Unreinforced and Reinforced Plastics and Electrical Insulating Materials<sup>1</sup>

This standard is issued under the fixed designation D 790; the number immediately following the designation indicates the year of original adoption or, in the case of revision, the year of last revision. A number in parentheses indicates the year of last reapproval. A superscript epsilon ( $\epsilon$ ) indicates an editorial change since the last revision or reapproval.

*This standard has been approved for use by agencies of the Department of Defense.*

### 1. Scope\*

1.1 These test methods cover the determination of flexural properties of unreinforced and reinforced plastics, including high-modulus composites and electrical insulating materials in the form of rectangular bars molded directly or cut from sheets, plates, or molded shapes. These test methods are generally applicable to both rigid and semirigid materials. However, flexural strength cannot be determined for those materials that do not break or that do not fail in the outer surface of the test specimen within the 5.0 % strain limit of these test methods. These test methods utilize a three-point loading system applied to a simply supported beam. A four-point loading system method can be found in Test Method D 6272.

1.1.1 *Procedure A*, designed principally for materials that break at comparatively small deflections.

1.1.2 *Procedure B*, designed particularly for those materials that undergo large deflections during testing.

1.1.3 Procedure A shall be used for measurement of flexural properties, particularly flexural modulus, unless the material specification states otherwise. Procedure B may be used for measurement of flexural strength only. Tangent modulus data obtained by Procedure A tends to exhibit lower standard deviations than comparable data obtained by means of Procedure B.

1.2 Comparative tests may be run in accordance with either procedure, provided that the procedure is found satisfactory for the material being tested.

1.3 The values stated in SI units are to be regarded as the standard. The values provided in parentheses are for information only.

1.4 *This standard does not purport to address all of the safety concerns, if any, associated with its use. It is the responsibility of the user of this standard to establish appropriate safety and health practices and determine the applicability of regulatory limitations prior to use.*

NOTE 1—These test methods are not technically equivalent to ISO 178.

### 2. Referenced Documents

2.1 *ASTM Standards:*

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<sup>1</sup> These test methods are under the jurisdiction of ASTM Committee D20 on Plastics and are the direct responsibility of Subcommittee D20.10 on Mechanical Properties. Current edition approved April 10, 2023. Published June 2023. Originally published as D 790—70; approved in 1970. Last previous edition approved in 2002 as D 790 – 002.

\*A Summary of Changes section appears at the end of this standard.

- D 618 Practice for Conditioning Plastics for Testing<sup>2</sup>
- D 638 Test Method for Tensile Properties of Plastics<sup>2</sup>
- D 883 Terminology Relating to Plastics<sup>2</sup>
- D 4000 Classification System for Specifying Plastic Materials<sup>3</sup>
- D 5947 Test Methods for Physical Dimensions of Solid Plastic Specimens<sup>4</sup>
- D 6272 Test Method for Flexural Properties of Unreinforced and Reinforced Plastics and Electrical Insulating Materials by Four-Point Bending<sup>4</sup>
- E 4 Practices for Force Verification of Testing Machines<sup>5</sup>
- E 691 Practice for Conducting an Interlaboratory Study to Determine the Precision of a Test Method<sup>6</sup>

**3. Terminology**

3.1 *Definitions*—Definitions of terms applying to these test methods appear in Terminology D 883 and Annex A1 of Test Method D 638.

**4. Summary of Test Method**

4.1 A bar of rectangular cross section rests on two supports and is loaded by means of a loading nose midway between the supports (see Fig. 1). A support span-to-depth ratio of 16:1 shall be used unless there is reason to suspect that a larger span-to-depth ratio may be required, as may be the case for certain laminated materials (see Section 7 and Note 8 for guidance).

4.2 The specimen is deflected until rupture occurs in the outer surface of the test specimen or until a maximum strain (see 12.7) of 5.0 % is reached, whichever occurs first.

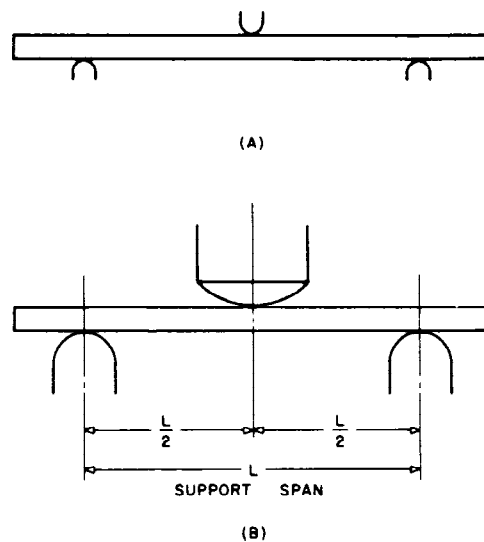
4.3 Procedure A employs a strain rate of 0.01 mm/mm/min ( [0.01 in./in./min] ) and is the preferred procedure for this test method, while Procedure B employs a strain rate of 0.10 mm/mm/min ( [0.10 in./in./min] ).

**5. Significance and Use**

5.1 Flexural properties as determined by these test methods are especially useful for quality control and specification purposes.

5.2 Materials that do not fail by the maximum strain allowed under these test methods (3-point bend) may be more suited to a 4-point bend test. The basic difference between the two test methods is in the location of the maximum bending moment and maximum axial fiber stresses. The maximum axial fiber stresses occur on a line under the loading nose in 3-point bending and over the area between the loading noses in 4-point bending.

<sup>2</sup> Annual Book of ASTM Standards, Vol 08.01.  
<sup>3</sup> Annual Book of ASTM Standards, Vol 08.02.  
<sup>4</sup> Annual Book of ASTM Standards, Vol 08.03.  
<sup>5</sup> Annual Book of ASTM Standards, Vol 03.01.  
<sup>6</sup> Annual Book of ASTM Standards, Vol 14.02.



NOTE—(a) Minimum radius = 3.2 mm ( [1/8 in.] ). (b) Maximum radius supports 1.6 times specimen depth; maximum radius loading nose = 4 times specimen depth.

**FIG. 1 Allowable Range of Loading Nose and Support Radii**

5.3 Flexural properties may vary with specimen depth, temperature, atmospheric conditions, and the difference in rate of straining as specified in Procedures A and B (see also Note 8).

5.4 Before proceeding with these test methods, reference should be made to the specification of the material being tested. Any test specimen preparation, conditioning, dimensions, or testing parameters, or combination thereof, covered in the materials specification shall take precedence over those mentioned in these test methods. If there are no material specifications, then the default conditions apply. Table 1 in Classification System D 4000 lists the ASTM materials standards that currently exist for plastics.

## 6. Apparatus

6.1 *Testing Machine*— A properly calibrated testing machine that can be operated at constant rates of crosshead motion over the range indicated, and in which the error in the load measuring system shall not exceed  $\pm 1\%$  of the maximum load expected to be measured. It shall be equipped with a deflection measuring device. The stiffness of the testing machine shall be such that the total elastic deformation of the system does not exceed  $1\%$  of the total deflection of the test specimen during testing, or appropriate corrections shall be made. The load indicating mechanism shall be essentially free from inertial lag at the crosshead rate used. The accuracy of the testing machine shall be verified in accordance with Practices E 4.

6.2 *Loading Noses and Supports* —The loading nose and supports shall have cylindrical surfaces. In order to avoid excessive indentation, or failure due to stress concentration directly under the loading nose, the radii of the loading nose and supports shall be  $5.0 \pm 0.1$  mm—([0.197  $\pm$  0.004 in.]) unless otherwise specified or agreed upon between the interested clients. When other loading noses and supports are used they must comply with the following requirements: they shall have a minimum radius of 3.2 mm—([ $\frac{1}{8}$  in.]) for all specimens, and for specimens 3.2 mm or greater in depth, the radius of the supports may be up to 1.6 times the specimen depth. They shall be this large if significant indentation or compressive failure occurs. The arc of the loading nose in contact with the specimen shall be sufficiently large to prevent contact of the specimen with the sides of the nose (see Fig. 1). The maximum radius of the loading nose shall be no more than 4 times the specimen depth.

NOTE 2—Test data have shown that the loading nose and support dimensions can influence the flexural modulus and flexural strength values. The loading nose dimension has the greater influence. Dimensions of the loading nose and supports must be specified in the material specification.

6.3 *Micrometers*— Suitable micrometers for measuring the width and thickness of the test specimen to an incremental discrimination of at least 0.025 mm—([0.001 in.]) should be used. All width and thickness measurements of rigid and semirigid plastics may be measured with a hand micrometer with ratchet. A suitable instrument for measuring the thickness of nonrigid test specimens shall have: a contact measuring pressure of  $25 \pm 2.5$  kPa—([3.6  $\pm$  0.36 psi]), a movable circular contact foot  $6.35 \pm 0.025$  mm—([0.250  $\pm$  0.001 in.]) in diameter and a lower fixed anvil large enough to extend beyond the contact foot in all directions and being parallel to the contact foot within 0.005 mm—([0.002 in.]) over the entire foot area. Flatness of foot and anvil shall conform to the portion of the Calibration section of Test Methods D 5947.

## 7. Test Specimens

7.1 The specimens may be cut from sheets, plates, or molded shapes, or may be molded to the desired finished dimensions. The actual dimensions used in Section 4.2, Calculation, shall be measured in accordance with Test Methods D 5947.

NOTE 3—Any necessary polishing of specimens shall be done only in the lengthwise direction of the specimen.

7.2 *Sheet Materials (Except Laminated Thermosetting Materials and Certain Materials Used for Electrical Insulation, Including Vulcanized Fiber and Glass Bonded Mica):*

7.2.1 *Materials 1.6 mm—([ $\frac{1}{16}$  in.]) or Greater in Thickness*—For flatwise tests, the depth of the specimen shall be the thickness of the material. For edgewise tests, the width of the specimen shall be the thickness of the sheet, and the depth shall not exceed

**TABLE 1 Flexural Strength**

Material	Mean, $10^3$ psi	Values Expressed in Units of % of $10^3$ psi			
		$V_r^A$	$V_R^B$	$r^C$	$R^D$
ABS	9.99	1.59	6.05	4.44	17.2
DAP thermoset	14.3	6.58	6.58	18.6	18.6
Cast acrylic	16.3	1.67	11.3	4.73	32.0
GR polyester	19.5	1.43	2.14	4.05	6.08
GR polycarbonate	21.0	5.16	6.05	14.6	17.1
SMC	26.0	4.76	7.19	13.5	20.4

<sup>A</sup>  $V_r$  = within-laboratory coefficient of variation for the indicated material. It is obtained by first pooling the within-laboratory standard deviations of the test results from all of the participating laboratories:  $S_r = [((s_1)^2 + (s_2)^2 \dots + (s_n)^2)/n]$  <sup>1/2</sup> then  $V_r = (S_r \text{ divided by the overall average for the material}) \times 100$ .

<sup>B</sup>  $V_R$  = between-laboratory reproducibility, expressed as the coefficient of variation:  $S_R = (S_r^2 + S_L^2)^{1/2}$  where  $S_L$  is the standard deviation of laboratory means. Then:  $V_R = (S_R \text{ divided by the overall average for the material}) \times 100$ .

<sup>C</sup>  $r$  = within-laboratory critical interval between two test results =  $2.8 \times V_r$ .

<sup>D</sup>  $R$  = between-laboratory critical interval between two test results =  $2.8 \times V_R$ .

the width (see Notes 4 and 5). For all tests, the support span shall be 16 (tolerance  $\pm 1$ ) times the depth of the beam. Specimen width shall not exceed one fourth of the support span for specimens greater than 3.2 mm ( $\frac{1}{8}$  in.) in depth. Specimens 3.2 mm or less in depth shall be 12.7 mm ( $\frac{1}{2}$  in.) in width. The specimen shall be long enough to allow for overhanging on each end of at least 10 % of the support span, but in no case less than 6.4 mm ( $\frac{1}{4}$  in.) on each end. Overhang shall be sufficient to prevent the specimen from slipping through the supports.

NOTE 4—Whenever possible, the original surface of the sheet shall be unaltered. However, where testing machine limitations make it impossible to follow the above criterion on the unaltered sheet, one or both surfaces shall be machined to provide the desired dimensions, and the location of the specimens with reference to the total depth shall be noted. The value obtained on specimens with machined surfaces may differ from those obtained on specimens with original surfaces. Consequently, any specifications for flexural properties on thicker sheets must state whether the original surfaces are to be retained or not. When only one surface was machined, it must be stated whether the machined surface was on the tension or compression side of the beam.

NOTE 5—Edgewise tests are not applicable for sheets that are so thin that specimens meeting these requirements cannot be cut. If specimen depth exceeds the width, buckling may occur.

**7.2.2 Materials Less than 1.6 mm ( $\frac{1}{16}$  in.) in Thickness**—The specimen shall be 50.8 mm (~~2 in.~~) [2 in.] long by 12.7 mm ( $\frac{1}{2}$  in.) wide, tested flatwise on a 25.4-mm (~~1 in.~~) [1 in.] support span.

NOTE 6—Use of the formulas for simple beams cited in these test methods for calculating results presumes that beam width is small in comparison with the support span. Therefore, the formulas do not apply rigorously to these dimensions.

NOTE 7—Where machine sensitivity is such that specimens of these dimensions cannot be measured, wider specimens or shorter support spans, or both, may be used, provided the support span-to-depth ratio is at least 14 to 1. All dimensions must be stated in the report (see also Note 6).

**7.3 Laminated Thermosetting Materials and Sheet and Plate Materials Used for Electrical Insulation, Including Vulcanized Fiber and Glass-Bonded Mica**—For paper-base and fabric-base grades over 25.4 mm (~~1 in.~~) [1 in.] in nominal thickness, the specimens shall be machined on both surfaces to a depth of 25.4 mm. For glass-base and nylon-base grades, specimens over 12.7 mm ( $\frac{1}{2}$  in.) in nominal depth shall be machined on both surfaces to a depth of 12.7 mm. The support span-to-depth ratio shall be chosen such that failures occur in the outer fibers of the specimens, due only to the bending moment (see Note 8). Therefore, a ratio larger than 16:1 may be necessary (32:1 or 40:1 are recommended). When laminated materials exhibit low compressive strength perpendicular to the laminations, they shall be loaded with a large radius loading nose (up to four times the specimen depth to prevent premature damage to the outer fibers).

**7.4 Molding Materials (Thermoplastics and Thermosets)**—The recommended specimen for molding materials is 127 by 12.7 by 3.2 mm (~~5~~ [5 by  $\frac{1}{2}$  by  $\frac{1}{8}$  in.]) tested flatwise on a support span, resulting in a support span-to-depth ratio of 16 (tolerance  $\pm 1$ ). Thicker specimens should be avoided if they exhibit significant shrink marks or bubbles when molded.

**7.5 High-Strength Reinforced Composites, Including Highly Orthotropic Laminates**—The span-to-depth ratio shall be chosen such that failure occurs in the outer fibers of the specimens and is due only to the bending moment (see Note 8). A span-to-depth ratio larger than 16:1 may be necessary (32:1 or 40:1 are recommended). For some highly anisotropic composites, shear deformation can significantly influence modulus measurements, even at span-to-depth ratios as high as 40:1. Hence, for these materials, an increase in the span-to-depth ratio to 60:1 is recommended to eliminate shear effects when modulus data are required, it should also be noted that the flexural modulus of highly anisotropic laminates is a strong function of ply-stacking sequence and will not necessarily correlate with tensile modulus, which is not stacking-sequence dependent.

NOTE 8—As a general rule, support span-to-depth ratios of 16:1 are satisfactory when the ratio of the tensile strength to shear strength is less than 8 to 1, but the support span-to-depth ratio must be increased for composite laminates having relatively low shear strength in the plane of the laminate and relatively high tensile strength parallel to the support span.

## 8. Number of Test Specimens

8.1 Test at least five specimens for each sample in the case of isotropic materials or molded specimens.

8.2 For each sample of anisotropic material in sheet form, test at least five specimens for each of the following conditions. Recommended conditions are flatwise and edgewise tests on specimens cut in lengthwise and crosswise directions of the sheet. For the purposes of this test, “lengthwise” designates the principal axis of anisotropy and shall be interpreted to mean the direction of the sheet known to be stronger in flexure. “Crosswise” indicates the sheet direction known to be the weaker in flexure and shall be at 90° to the lengthwise direction.

## 9. Conditioning

9.1 **Conditioning**—Condition the test specimens at  $23 \pm 2^\circ\text{C}$  ( $[73.4 \pm 3.6^\circ\text{F}]$ ) and  $50 \pm 5\%$  relative humidity for not less than 40 h prior to test in accordance with Procedure A of Practice D 618 unless otherwise specified by contract or the relevant ASTM material specification. Reference pre-test conditioning, to settle disagreements, shall apply tolerances of  $\pm 1^\circ\text{C}$  ( $\pm 1.8^\circ\text{F}$ ) and  $\pm 2\%$  relative humidity.

9.2 **Test Conditions**—Conduct the tests at  $23 \pm 2^\circ\text{C}$  ( $[73.4 \pm 3.6^\circ\text{F}]$ ) and  $50 \pm 5\%$  relative humidity unless otherwise specified by contract or the relevant ASTM material specification. Reference testing conditions, to settle disagreements, shall apply tolerances of  $\pm 1^\circ\text{C}$  ( $\pm 1.8^\circ\text{F}$ ) [1.8°F] and  $\pm 2\%$  relative humidity.

## 10. Procedure

10.1 **Procedure A:**

10.1.1 Use an untested specimen for each measurement. Measure the width and depth of the specimen to the nearest 0.03 mm ([0.001 in.]) at the center of the support span. For specimens less than 2.54 mm ([0.100 in.]) in depth, measure the depth to the nearest 0.003 mm ([0.0005 in.]). These measurements shall be made in accordance with Test Methods D 5947.

10.1.2 Determine the support span to be used as described in Section 7 and set the support span to within 1 % of the determined value.

10.1.3 For flexural fixtures that have continuously adjustable spans, measure the span accurately to the nearest 0.1 mm ([0.004 in.]) for spans less than 63 mm ([2.5 in.]) and to the nearest 0.3 mm ([0.012 in.]) for spans greater than or equal to 63 mm ([2.5 in.]). Use the actual measured span for all calculations. For flexural fixtures that have fixed machined span positions, verify the span distance the same as for adjustable spans at each machined position. This distance becomes the span for that position and is used for calculations applicable to all subsequent tests conducted at that position. See Annex A2 for information on the determination of and setting of the span.

10.1.4 Calculate the rate of crosshead motion as follows and set the machine for the rate of crosshead motion as calculated by Eq 1:

$$R = ZL^2/6d \quad (1)$$

where:

$R$  = rate of crosshead motion, mm-(in.)/min,

$L$  = support span, mm-(in.); [in.],

$d$  = depth of beam, mm-(in.); [in.], and

$Z$  = rate of straining of the outer fiber, mm/mm/min-[in./in./min)].  $Z$  shall be equal to 0.01.

In no case shall the actual crosshead rate differ from that calculated using Eq 1, by more than  $\pm 10$  %.

10.1.5 Align the loading nose and supports so that the axes of the cylindrical surfaces are parallel and the loading nose is midway between the supports. The parallelism of the apparatus may be checked by means of a plate with parallel grooves into which the loading nose and supports will fit when properly aligned (see A2.3). Center the specimen on the supports, with the long axis of the specimen perpendicular to the loading nose and supports.

10.1.6 Apply the load to the specimen at the specified crosshead rate, and take simultaneous load-deflection data. Measure deflection either by a gage under the specimen in contact with it at the center of the support span, the gage being mounted stationary relative to the specimen supports, or by measurement of the motion of the loading nose relative to the supports. Load-deflection curves may be plotted to determine the flexural strength, chord or secant modulus or the tangent modulus of elasticity, and the total work as measured by the area under the load-deflection curve. Perform the necessary toe compensation (see Annex A1) to correct for seating and indentation of the specimen and deflections in the machine.

10.1.7 Terminate the test when the maximum strain in the outer surface of the test specimen has reached 0.05 mm/mm-[in./in.] or at break if break occurs prior to reaching the maximum strain (Notes 9 and 10). The deflection at which this strain will occur may be calculated by letting  $r$  equal 0.05 mm/mm-[in./in.] in Eq 2:

$$D = rL^2/6d \quad (2)$$

where:

$D$  = midspan deflection, mm-(in.); [in.],

$r$  = strain, mm/mm-[in./in.],

$L$  = support span, mm-(in.); [in.], and

$d$  = depth of beam, mm-(in.); [in.].

NOTE 9—For some materials that do not yield or break within the 5 % strain limit when tested by Procedure A, the increased strain rate allowed by Procedure B (see 10.2) may induce the specimen to yield or break, or both, within the required 5 % strain limit.

NOTE 10—Beyond 5 % strain, this test method is not applicable. Some other mechanical property might be more relevant to characterize materials that neither yield nor break by either Procedure A or Procedure B within the 5 % strain limit (for example, Test Method D 638 may be considered).

## 10.2 Procedure B:

10.2.1 Use an untested specimen for each measurement.

10.2.2 Test conditions shall be identical to those described in 10.1, except that the rate of straining of the outer surface of the test specimen shall be 0.10 mm/mm-[in./in.]/min.

10.2.3 If no break has occurred in the specimen by the time the maximum strain in the outer surface of the test specimen has reached 0.05 mm/mm-[in./in.], discontinue the test (see Note 10).

## 11. Retests

11.1 Values for properties at rupture shall not be calculated for any specimen that breaks at some obvious, fortuitous flaw, unless such flaws constitute a variable being studied. Retests shall be made for any specimen on which values are not calculated.

## 12. Calculation

12.1 Toe compensation shall be made in accordance with Annex A1 unless it can be shown that the toe region of the curve is not due to the take-up of slack, seating of the specimen, or other artifact, but rather is an authentic material response.

12.2 *Flexural Stress* ( $\sigma_f$ )—When a homogeneous elastic material is tested in flexure as a simple beam supported at two points and loaded at the midpoint, the maximum stress in the outer surface of the test specimen occurs at the midpoint. This stress may be calculated for any point on the load-deflection curve by means of the following equation (see Notes 11-13):

$$\sigma_f = 3PL/2bd^2 \quad (3)$$

where:

- $\sigma$  = stress in the outer fibers at midpoint, MPa-(psi)-[psi],
- $P$  = load at a given point on the load-deflection curve, N-(lbf)-[lbf],
- $L$  = support span, mm-(in.)-[in.],
- $b$  = width of beam tested, mm-(in.)-[in.], and
- $d$  = depth of beam tested, mm-(in.)-[in.].

NOTE 11—Eq 3 applies strictly to materials for which stress is linearly proportional to strain up to the point of rupture and for which the strains are small. Since this is not always the case, a slight error will be introduced if Eq 3 is used to calculate stress for materials that are not true Hookean materials. The equation is valid for obtaining comparison data and for specification purposes, but only up to a maximum fiber strain of 5 % in the outer surface of the test specimen for specimens tested by the procedures described herein.

NOTE 12—When testing highly orthotropic laminates, the maximum stress may not always occur in the outer surface of the test specimen.<sup>7</sup> Laminated beam theory must be applied to determine the maximum tensile stress at failure. If Eq 3 is used to calculate stress, it will yield an apparent strength based on homogeneous beam theory. This apparent strength is highly dependent on the ply-stacking sequence of highly orthotropic laminates.

NOTE 13—The preceding calculation is not valid if the specimen slips excessively between the supports.

12.3 *Flexural Stress for Beams Tested at Large Support Spans* ( $\sigma_f$ )—If support span-to-depth ratios greater than 16 to 1 are used such that deflections in excess of 10 % of the support span occur, the stress in the outer surface of the specimen for a simple beam can be reasonably approximated with the following equation (see Note 14):

$$\sigma_f = (3PL/2bd^2)[1 + 6(D/L)^2 - 4(d/L)(D/L)] \quad (4)$$

where:

$\sigma_f$ ,  $P$ ,  $L$ ,  $b$ , and  $d$  are the same as for Eq 3, and

$D$  = deflection of the centerline of the specimen at the middle of the support span, mm-(in.)-[in.].

NOTE 14—When large support span-to-depth ratios are used, significant end forces are developed at the support noses which will affect the moment in a simple supported beam. Eq 4 includes additional terms that are an approximate correction factor for the influence of these end forces in large support span-to-depth ratio beams where relatively large deflections exist.

12.4 *Flexural Strength* ( $\sigma_{fM}$ )—Maximum flexural stress sustained by the test specimen (see Note 12) during a bending test. It is calculated according to Eq 3 or Eq 4. Some materials that do not break at strains of up to 5 % may give a load deflection curve that shows a point at which the load does not increase with an increase in strain, that is, a yield point (Fig. 2, Curve B),  $Y$ . The flexural strength may be calculated for these materials by letting  $P$  (in Eq 3 or Eq 4) equal this point,  $Y$ .

12.5 *Flexural Offset Yield Strength*—Offset yield strength is the stress at which the stress-strain curve deviates by a given strain (offset) from the tangent to the initial straight line portion of the stress-strain curve. The value of the offset must be given whenever this property is calculated.

NOTE 15—This value may differ from flexural strength defined in 12.4. Both methods of calculation are described in the annex to Test Method D 638.

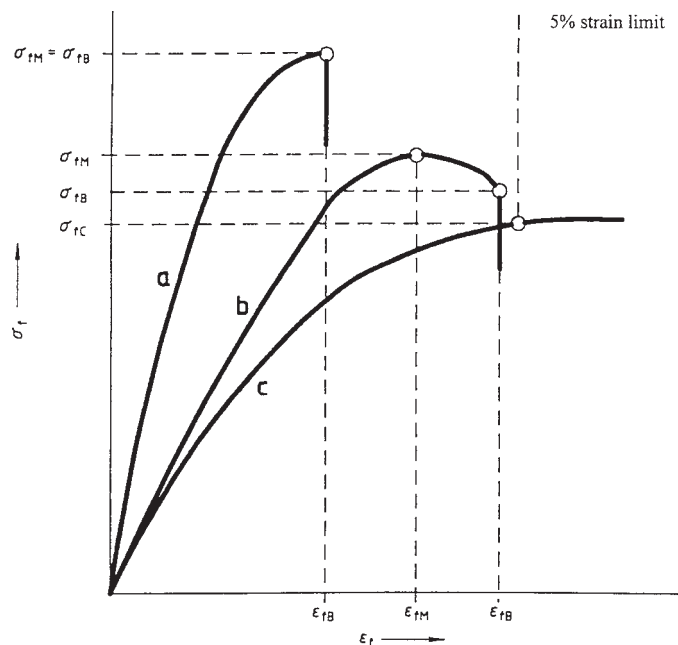
12.6 *Flexural Stress at Break* ( $\sigma_{fB}$ )—Flexural stress at break of the test specimen during a bending test. It is calculated according to Eq 3 or Eq 4. Some materials may give a load deflection curve that shows a break point,  $B$ , without a yield point (Fig. 2, Curve a) in which case  $\sigma_{fB} = \sigma_{fM}$ . Other materials may give a yield deflection curve with both a yield and a break point,  $B$  (Fig. 2, Curve b). The flexural stress at break may be calculated for these materials by letting  $P$  (in Eq 3 or Eq 4) equal this point,  $B$ .

12.7 *Stress at a Given Strain*—The stress in the outer surface of a test specimen at a given strain may be calculated in accordance with Eq 3 or Eq 4 by letting  $P$  equal the load read from the load-deflection curve at the deflection corresponding to the desired strain (for highly orthotropic laminates, see Note 12).

12.8 *Flexural Strain*,  $\epsilon_f$ —Nominal fractional change in the length of an element of the outer surface of the test specimen at midspan, where the maximum strain occurs. It may be calculated for any deflection using Eq 5:

$$\epsilon_f = 6Dd/L^2 \quad (5)$$

<sup>7</sup> For a discussion of these effects, see Zweben, C., Smith, W. S., and Wardle, M. W., "Test Methods for Fiber Tensile Strength, Composite Flexural Modulus and Properties of Fabric-Reinforced Laminates," *Composite Materials: Testing and Design (Fifth Conference)*, ASTM STP 674, 1979, pp. 228–262.



NOTE—Curve a: Specimen that breaks before yielding.  
 Curve b: Specimen that yields and then breaks before the 5 % strain limit.  
 Curve c: Specimen that neither yields nor breaks before the 5 % strain limit.

**FIG. 2 Typical Curves of Flexural Stress ( $\sigma_f$ ) Versus Flexural Strain ( $\epsilon_f$ )**

where:

- $\epsilon_f$  = strain in the outer surface, mm/mm-([in./in.]),
- $D$  = maximum deflection of the center of the beam, mm-(in.), [in.],
- $L$  = support span, mm-(in.), [in.], and
- $d$  = depth, mm-(in.), [in.],
- $D$  = maximum deflection of the center of the beam, mm-(in.), [in.],
- $L$  = support span, mm-(in.), [in.], and
- $d$  = depth, mm-(in.), [in.].

12.9 Modulus of Elasticity:

12.9.1 *Tangent Modulus of Elasticity*—The tangent modulus of elasticity, often called the “modulus of elasticity,” is the ratio, within the elastic limit, of stress to corresponding strain. It is calculated by drawing a tangent to the steepest initial straight-line portion of the load-deflection curve and using Eq 6 (for highly anisotropic composites, see Note 16).

$$E_B = L^3m/4bd^3 \tag{6}$$

where:

- $E_B$  = modulus of elasticity in bending, MPa-(psi), [psi],
- $L$  = support span, mm-(in.), [in.],
- $b$  = width of beam tested, mm-(in.), [in.],
- $d$  = depth of beam tested, mm-(in.), [in.], and
- $m$  = slope of the tangent to the initial straight-line portion of the load-deflection curve, N/mm-([lbf/in.]) of deflection.

NOTE 16—Shear deflections can seriously reduce the apparent modulus of highly anisotropic composites when they are tested at low span-to-depth ratios.<sup>7</sup> For this reason, a span-to-depth ratio of 60 to 1 is recommended for flexural modulus determinations on these composites. Flexural strength should be determined on a separate set of replicate specimens at a lower span-to-depth ratio that induces tensile failure in the outer fibers of the beam along its lower face. Since the flexural modulus of highly anisotropic laminates is a critical function of ply-stacking sequence, it will not necessarily correlate with tensile modulus, which is not stacking-sequence dependent.

12.9.2 *Secant Modulus*— The secant modulus is the ratio of stress to corresponding strain at any selected point on the stress-strain curve, that is, the slope of the straight line that joins the origin and a selected point on the actual stress-strain curve. It shall be expressed in megapascals-([pounds per square inch]). The selected point is chosen at a prespecified stress or strain in accordance with the appropriate material specification or by customer contract. It is calculated in accordance with Eq 6 by letting

**TABLE 2 Flexural Modulus**

Material	Mean, 10 <sup>3</sup> psi	Values Expressed in units of % of 10 <sup>3</sup> psi			
		V <sub>r</sub> <sup>A</sup>	V <sub>R</sub> <sup>B</sup>	r <sup>C</sup>	R <sup>D</sup>
ABS	338	4.79	7.69	13.6	21.8
DAP thermoset	485	2.89	7.18	8.15	20.4
Cast acrylic	810	13.7	16.1	38.8	45.4
GR polyester	816	3.49	4.20	9.91	11.9
GR polycarbonate	1790	5.52	5.52	15.6	15.6
SMC	1950	10.9	13.8	30.8	39.1

<sup>A</sup> V<sub>r</sub> = within-laboratory coefficient of variation for the indicated material. It is obtained by first pooling the within-laboratory standard deviations of the test results from all of the participating laboratories:  $S_r = [((s_1)^2 + (s_2)^2 \dots + (s_n)^2)/n]$ <sup>1/2</sup> then V<sub>r</sub> = (S<sub>r</sub> divided by the overall average for the material) × 100.

<sup>B</sup> V<sub>R</sub> = between-laboratory reproducibility, expressed as the coefficient of variation:  $S_R = \{S_r^2 + S_L^2\}^{1/2}$  where S<sub>L</sub> is the standard deviation of laboratory means. Then: V<sub>R</sub> = (S<sub>R</sub> divided by the overall average for the material) × 100.

<sup>C</sup> r = within-laboratory critical interval between two test results = 2.8 × V<sub>r</sub>.

<sup>D</sup> R = between-laboratory critical interval between two test results = 2.8 × V<sub>R</sub>.

*m* equal the slope of the secant to the load-deflection curve. The chosen stress or strain point used for the determination of the secant shall be reported.

**12.9.3 Chord Modulus (E<sub>f</sub>)**—The chord modulus may be calculated from two discrete points on the load deflection curve. The selected points are to be chosen at two prespecified stress or strain points in accordance with the appropriate material specification or by customer contract. The chosen stress or strain points used for the determination of the chord modulus shall be reported. Calculate the chord modulus, E<sub>f</sub> using the following equation:

$$E_f = (\sigma_{f2} - \sigma_{f1}) / (\epsilon_{f2} - \epsilon_{f1}) \quad (7)$$

where:

σ<sub>f2</sub> and σ<sub>f1</sub> are the flexural stresses, calculated from Eq 3 or Eq 4 and measured at the predefined points on the load deflection curve, and ε<sub>f2</sub> and

ε<sub>f1</sub> are the flexural strain values, calculated from Eq 5 and measured at the predetermined points on the load deflection curve.

**12.10 Arithmetic Mean**—For each series of tests, the arithmetic mean of all values obtained shall be calculated to three significant figures and reported as the “average value” for the particular property in question.

**12.11 Standard Deviation**—The standard deviation (estimated) shall be calculated as follows and be reported to two significant figures:

$$s = \sqrt{(\sum X^2 - n\bar{X}^2) / (n - 1)} \quad (8)$$

where:

*s* = estimated standard deviation,

*X* = value of single observation,

*n* = number of observations, and

$\bar{X}$  = arithmetic mean of the set of observations.

### 13. Report

13.1 Report the following information:

13.1.1 Complete identification of the material tested, including type, source, manufacturer’s code number, form, principal dimensions, and previous history (for laminated materials, ply-stacking sequence shall be reported),

13.1.2 Direction of cutting and loading specimens, when appropriate,

13.1.3 Conditioning procedure,

13.1.4 Depth and width of specimen,

13.1.5 Procedure used (A or B),

13.1.6 Support span length,

13.1.7 Support span-to-depth ratio if different than 16:1,

13.1.8 Radius of supports and loading noses if different than 5 mm,

13.1.9 Rate of crosshead motion,

13.1.10 Flexural strain at any given stress, average value and standard deviation,

13.1.11 If a specimen is rejected, reason(s) for rejection,

13.1.12 Tangent, secant, or chord modulus in bending, average value, standard deviation, and the strain level(s) used if secant or chord modulus,

13.1.13 Flexural strength (if desired), average value, and standard deviation,



- 13.1.14 Stress at any given strain up to and including 5 % (if desired), with strain used, average value, and standard deviation,
- 13.1.15 Flexural stress at break (if desired), average value, and standard deviation,
- 13.1.16 Type of behavior, whether yielding or rupture, or both, or other observations, occurring within the 5 % strain limit, and
- 13.1.17 Date of specific version of test used.

#### 14. Precision and Bias <sup>8</sup>

14.1 Tables 1 and 2 are based on a round-robin test conducted in 1984, in accordance with Practice E 691, involving six materials tested by six laboratories using Procedure A. For each material, all the specimens were prepared at one source. Each “test result” was the average of five individual determinations. Each laboratory obtained two test results for each material.

NOTE 17—**Caution:** The following explanations of  $r$  and  $R$  (14.2-14.2.3) are intended only to present a meaningful way of considering the approximate precision of these test methods. The data given in Tables 2 and 3 should not be applied rigorously to the acceptance or rejection of materials, as those data are specific to the round robin and may not be representative of other lots, conditions, materials, or laboratories. Users of these test methods should apply the principles outlined in Practice E 691 to generate data specific to their laboratory and materials, or between specific laboratories. The principles of 14.2-14.2.3 would then be valid for such data.

14.2 *Concept of “r” and “R” in Tables 1 and 2*—If  $S_r$  and  $S_R$  have been calculated from a large enough body of data, and for test results that were averages from testing five specimens for each test result, then:

14.2.1 *Repeatability*— Two test results obtained within one laboratory shall be judged not equivalent if they differ by more than the  $r$  value for that material.  $r$  is the interval representing the critical difference between two test results for the same material, obtained by the same operator using the same equipment on the same day in the same laboratory.

14.2.2 *Reproducibility*— Two test results obtained by different laboratories shall be judged not equivalent if they differ by more than the  $R$  value for that material.  $R$  is the interval representing the critical difference between two test results for the same material, obtained by different operators using different equipment in different laboratories.

14.2.3 The judgments in 14.2.1 and 14.2.2 will have an approximately 95 % (0.95) probability of being correct.

14.3 *Bias*—No statement may be made about the bias of these test methods, as there is no standard reference material or reference test method that is applicable.

#### 15. Keywords

- 15.1 flexural properties; plastics; stiffness; strength

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<sup>8</sup> Supporting data are available from ASTM Headquarters. Request RR: D20 – 1128.

## ANNEXES

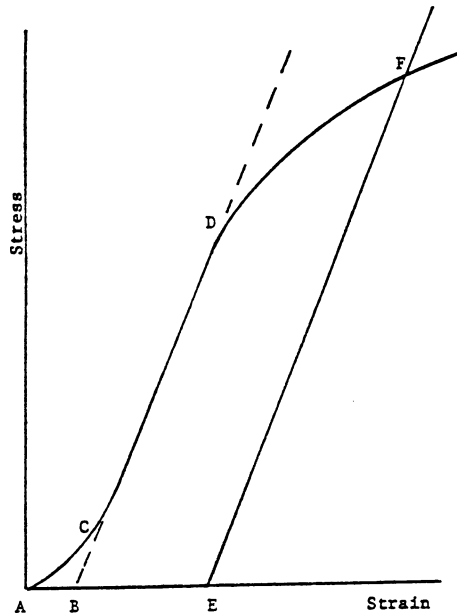
### (Mandatory Information)

#### A1. TOE COMPENSATION

A1.1 In a typical stress-strain curve (see Fig. A1.1) there is a toe region,  $AC$ , that does not represent a property of the material. It is an artifact caused by a takeup of slack and alignment or seating of the specimen. In order to obtain correct values of such parameters as modulus, strain, and offset yield point, this artifact must be compensated for to give the corrected zero point on the strain or extension axis.

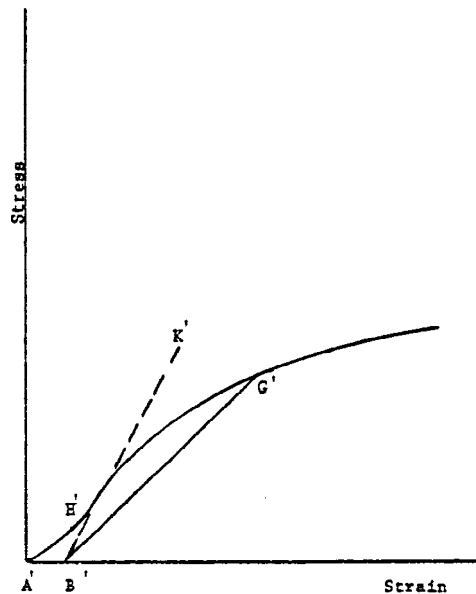
A1.2 In the case of a material exhibiting a region of Hookean (linear) behavior (see Fig. A1.1), a continuation of the linear ( $CD$ ) region of the curve is constructed through the zero-stress axis. This intersection ( $B$ ) is the corrected zero-strain point from which all extensions or strains must be measured, including the yield offset ( $BE$ ), if applicable. The elastic modulus can be determined by dividing the stress at any point along the Line  $CD$  (or its extension) by the strain at the same point (measured from Point  $B$ , defined as zero-strain).

A1.3 In the case of a material that does not exhibit any linear region (see Fig. A1.2), the same kind of toe correction of the zero-strain point can be made by constructing a tangent to the maximum slope at the inflection Point  $H'$ . This is extended to intersect the strain axis at Point  $B'$ , the corrected zero-strain point. Using Point  $B'$  as zero strain, the stress at any point ( $G'$ ) on the curve can be divided by the strain at that point to obtain a secant modulus (slope of Line  $B'G'$ ). For those materials with no linear region, any attempt to use the tangent through the inflection point as a basis for determination of an offset yield point may result in unacceptable error.



NOTE—Some chart recorders plot the mirror image of this graph.

**FIG. A1.1 Material with Hookean Region**



NOTE—Some chart recorders plot the mirror image of this graph.

**FIG. A1.2 Material with No Hookean Region**

**A2. MEASURING AND SETTING SPAN**

A2.1 For flexural fixtures that have adjustable spans, it is important that the span between the supports is maintained constant or the actual measured span is used in the calculation of stress, modulus, and strain, and the loading nose or noses are positioned and aligned properly with respect to the supports. Some simple steps as follows can improve the repeatability of your results when using these adjustable span fixtures.

**A2.2 Measurement of Span:**

A2.2.1 This technique is needed to ensure that the correct span, not an estimated span, is used in the calculation of results.

A2.2.2 Scribe a permanent line or mark at the exact center of the support where the specimen makes complete contact. The type of mark depends on whether the supports are fixed or rotatable (see Figs. A2.1 and A2.2).

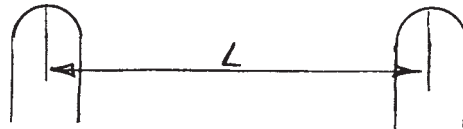


FIG. A2.1 Markings on Fixed Specimen Supports

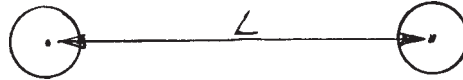


FIG. A2.2 Markings on Rotatable Specimen Supports

A2.2.3 Using a vernier caliper with pointed tips that is readable to at least 0.1 mm (0.004 in.), measure the distance between the supports, and use this measurement of span in the calculations.

A2.3 *Setting the Span and Alignment of Loading Nose(s)*—To ensure a consistent day-to-day setup of the span and ensure the alignment and proper positioning of the loading nose, simple jigs should be manufactured for each of the standard setups used. An example of a jig found to be useful is shown in Fig. A2.3.—

## APPENDIX

### (Nonmandatory Information)

#### X1. DEVELOPMENT OF A FLEXURAL MACHINE COMPLIANCE CORRECTION

##### X1.1 Introduction

X1.1.1 Universal Testing instrument drive systems always exhibit a certain level of compliance that is characterized by a variance between the reported crosshead displacement and the displacement actually imparted to the specimen. This variance is a function of load frame stiffness, drive system wind-up, load cell compliance and fixture compliance. To accurately measure the flexural modulus of a material, this compliance should be measured and empirically subtracted from test data. Flexural modulus results without the corrections are lower than if the correction is applied. The greater the stiffness of the material the more influence the system compliance has on results.

X1.1.2 It is not necessary to make the machine compliance correction when a deflectometer/extensometer is used to measure the actual deflection occurring in the specimen as it is deflected.

##### X1.2 Terminology

X1.2.1 *Compliance*—The displacement difference between test machine drive system displacement values and actual specimen displacement

X1.2.2 *Compliance Correction*—An analytical method of modifying test instrument displacement values to eliminate the amount of that measurement attributed to test instrument compliance.

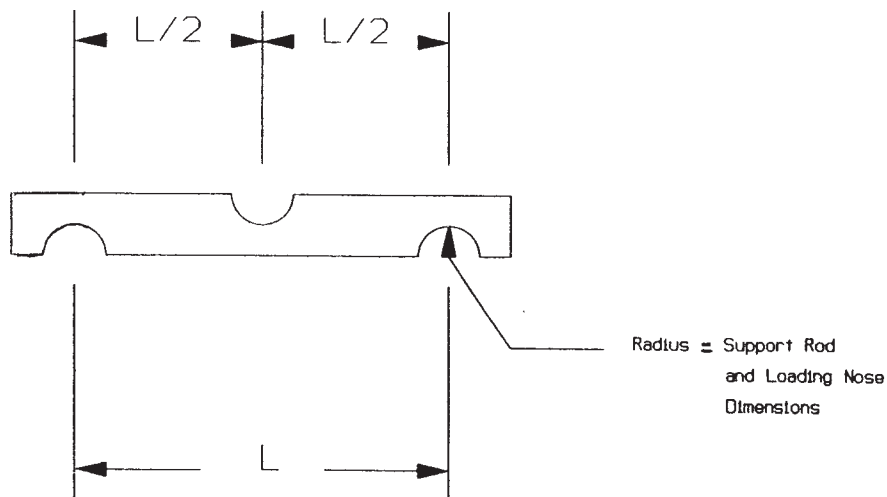


FIG. A2.3 Fixture Used to Set Loading Nose and Support Spacing and Alignment

### **X1.3 Apparatus**

X1.3.1 Universal Testing machine

X1.3.2 Load cell

X1.3.3 Flexure fixture including loading nose and specimen supports

X1.3.4 Computer Software to make corrections to the displacements

X1.3.5 Steel bar, with smoothed surfaces and a calculated flexural stiffness of more than 100 times greater than the test material. The length should be at least 13 mm greater than the support span. The width shall match the width of the test specimen and the thickness shall be that required to achieve or exceed the target stiffness.

### **X1.4 Safety Precautions**

X1.4.1 The universal testing machine should stop the machine crosshead movement when the load reaches 90 % of load cell capacity, to prevent damage to the load cell.

X1.4.2 The compliance curve determination should be made at a speed no higher than 2 mm/min. Because the load builds up rapidly since the steel bar does not deflect, it is quite easy to exceed the load cell capacity.

### **X1.5 Procedure**

NOTE X1.1—A new compliance correction curve should be established each time there is a change made to the setup of the test machine, such as, load cell changed or reinstallation of the flexure fixture on the machine. If the test machine is dedicated to flexural testing, and there are no changes to the setup, it is not necessary to re-calculate the compliance curve.

NOTE X1.2—On those machines with computer software that automatically make this compliance correction; refer to the software manual to determine how this correction should be made.

X1.5.1 The procedure to determine compliance follows:

X1.5.1.1 Configure the test system to match the actual test configuration.

X1.5.1.2 Place the steel bar in the test fixture, duplicating the position of a specimen during actual testing.

X1.5.1.3 Set the crosshead speed to 2 mm/min. or less and start the crosshead moving in the test direction recording crosshead displacement and the corresponding load values.

X1.5.1.4 Increase load to a point exceeding the highest load expected during specimen testing. Stop the crosshead and return to the pre-test location.

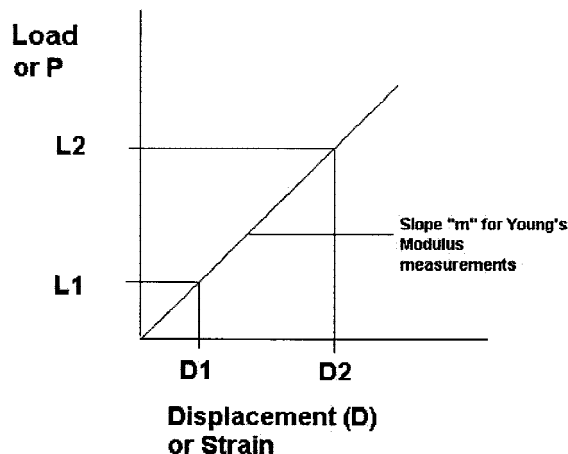
X1.5.1.5 The recorded load-deflection curve, starting when the loading nose contacts the steel bar to the time that the highest load expected is defined as test system compliance.

X1.5.2 Procedure to apply compliance correction is as follows:

X1.5.2.1 Run the flexural test method on the material at the crosshead required for the measurement.

X1.5.2.2 It is preferable that computer software be used to make the displacement corrections, but if it is not available compliance corrections can be made manually in the following manner. Determine the range of displacement (D) on the load versus displacement curve for the material, over which the modulus is to be calculated. For Young's Modulus that would steepest region of the curve below the proportional limit. For Secant and Chord Moduli that would be at specified level of strain or specified levels of strain, respectively. Draw two vertical lines up from the displacement axis for the two chosen displacements (D1, D2) to the load versus displacement curve for the material. In some cases one of these points maybe at zero displacement after the toe compensation correction is made. Draw two horizontal lines from these points on the load displacement curve to the Load (P) axis. Determine the loads (L1, L2).

X1.5.2.3 Using the Compliance Correction load displacement curve for the steel bar, mark off L1 and L2 on the Load (P) axis. From these two points draw horizontal lines across till they contact the load versus displacement curve for the steel bar. From these



**FIG. X1.1 Example of Modulus Curve for a Material**

two points on the load deflection curve draw two vertical lines downwards to the displacement axis. These two points on the displacement axis determine the corrections ( $c_1$ ,  $c_2$ ) that need to be made to the displacements measurements for the test material.

X1.5.2.4 Subtract the corrections ( $c_1$ ,  $c_2$ ) from the measured displacements ( $D_1$ ,  $D_2$ ), so that a true measures of test specimen deflection ( $D_1-c_1$ ,  $D_2-c_2$ ) are obtained.

**X1.6 Calculations**

X1.6.1 Calculation of Chord Modulus

X1.6.1.1 Calculate the stresses ( $\sigma_1$ ,  $\sigma_2$ ) for load points  $L_1$  and  $L_2$  from Fig. X1.1 using the equation in 12.2 3.

X1.6.1.2 Calculate the strains ( $\epsilon_1$ ,  $\epsilon_2$ ) for displacements  $D_1-c_1$  and  $D_2-c_2$  from Fig. X1.3 using the equation in 12.8 Eq. 5.

X1.6.1.3 Calculate the flexural chord modulus in accordance with 12.9.3 Eq. 7.

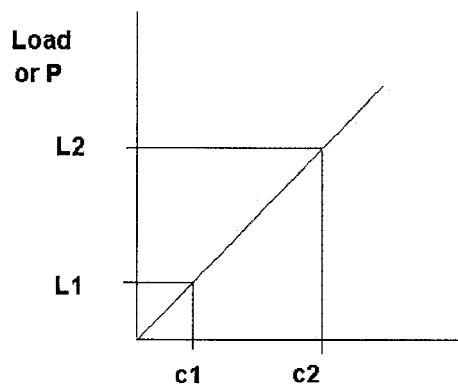
X1.6.2 Calculation of Secant Modulus

X1.6.2.1 Calculation of the Secant Modulus at any strain along the curve would be the same as conducting a chord modulus measurement, except that  $\sigma_1 = 0$ ,  $L_1 = 0$ , and  $D_1-c_1 = 0$ .

X1.6.3 Calculation of Young’s Modulus

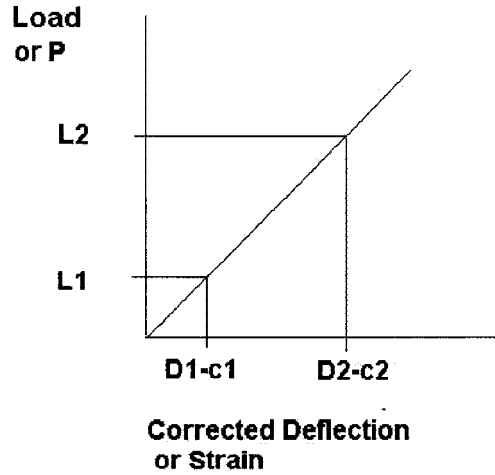
X1.6.3.1 Determine the steepest slope “ $m$ ” along the curve, below the proportional limit, using the selected loads  $L_1$  and  $L_2$  from Fig. X1.1 and the displacements  $D_1-c_1$  and  $D_2-c_2$  from Fig. X1.3.

X1.6.3.2 Calculate the Young’s modulus in accordance with 12.9.1 Eq. 6.



**Compliance Correction for Displacement (D) or Strain**

FIG. X1.2 Compliance Curve for Steel Bar



**FIG. X1.3 Example of the Material Curve Corrected for the Compliance Corrected Displacement or Strain**

### SUMMARY OF CHANGES

This section identifies the location of selected changes to these test methods. For the convenience of the user, Committee D20 has highlighted those changes that may impact the use of these test methods. This section may also include descriptions of the changes or reasons for the changes, or both.

D 790 – 03:

(I) Added Appendix X1.

D 790 – 02:

(I) Revised 9.1 and 9.2.

D 790 – 00:

(I) Revised 12.1.

D 790 – 99:

(I) Revised 10.1.3.

D 790 – 98:

- (1) Section 4.2 was rewritten extensively to bring this standard closer to ISO 178.
- (2) Fig. 2 was added to clarify flexural behaviors that may be observed and to define what yielding and breaking behaviors look like, as well as the appropriate place to select these points on the stress strain curve.

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